

**Reason**

(e) The mandatory continuing airworthiness information (MCAI) states:

Since 2004, more than 30 reports have been received of in-flight loss of a rear passenger door on Diamond aeroplanes, the majority of which were DA 40. In addition, at least 18 doors have been replaced because of damage found on the hinge.

Diamond Aircraft Industries conducted analyses and structural tests to determine the root cause of the door opening in flight. The conclusions were that the primary locking mechanism provided adequate strength to react to the loads in flight. It was also determined that the root cause was the crew not properly securing the rear passenger door by the main locking mechanism, prior to flight. Damage to the hinges has been caused primarily by external loads (wind gust conditions) while the aeroplane was parked.

All DA 40 and DA 42 aeroplanes have a system installed that provides a warning if the main door latch is not fully closed and a secondary safety latch (with retaining bracket) design feature. The initial intended design function of the latch was to hold the rear passenger door in the "near closed" position while on the ground, protecting the door from wind gusts. However, the original retaining bracket Part Number (P/N) DA4-5200-00-69 might not hold the door in this "near closed" position while in flight. To address this problem, DAI have designed an improved retaining bracket, P/N DA4-5200-00-69-SB, which has been satisfactory tested to hold the door closed in flight. In addition, DAI have revised the Airplane Flight Manual (AFM) emergency door unlocked/open procedure.

This condition, if not corrected, could result in the rear passenger door opening and departing the aeroplane in flight.

For the reasons described above, this AD requires implementation of amendment of the AFM procedures for flight with the door unlocked/open, and replacement of the passenger door retaining bracket with an improved part.

**Actions and Compliance**

(f) Unless already done, do the following actions:

(1) Within 6 months after the effective date of this AD, incorporate Diamond Aircraft Temporary Revision TR-MÄM 42-443, pages 3-55a and 3-55b, dated June 17, 2010, into the FAA-approved airplane flight manual following Diamond Aircraft Temporary Revision TR-MÄM 42-443, Cover Page, dated June 17, 2010.

(2) Within 6 months after the effective date of this AD, replace the rear passenger door retaining bracket with an improved design retaining bracket following Diamond Aircraft Industries GmbH Mandatory Service Bulletin No. MSB 42-083/No. MSB 42NG-014, dated July 13, 2010; and Diamond Aircraft Industries GmbH Work Instruction WI-MSB 42-083/WI-MSB 42NG-014, dated July 13, 2010.

(3) As of 6 months after the effective date of this AD, do not install a part number DA4-5200-00-69 rear passenger door retaining bracket.

**FAA AD Differences**

**Note:** This AD differs from the MCAI and/or service information as follows: On November 23, 2010, we issued AD 2010-25-01 as a unilateral action to address this unsafe condition on Diamond Aircraft Industries GmbH Models DA 40 and DA 40F airplanes. Subsequently, the European Aviation Safety Agency (EASA) issued AD 2010-0235 to address the same unsafe condition on both DA 40 and DA 42 series aeroplanes. Since AD 2010-25-01 already addresses this unsafe condition on Models DA 40 and DA 40F airplanes, we are not including those models in this AD.

**Other FAA AD Provisions**

(g) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Mike Kiesov, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4144; fax: (816) 329-4090. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, a Federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave., SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

**Related Information**

(h) Refer to MCAI EASA AD 2010-0235, dated November 10, 2010; Diamond Aircraft Industries GmbH Mandatory Service Bulletin No. MSB 42-083/No. MSB 42NG-014, dated July 13, 2010; Diamond Aircraft Industries GmbH Work Instruction WI-MSB 42-083/WI-MSB 42NG-014, dated July 13, 2010; and Diamond Aircraft Temporary Revision TR-

MÄM 42-443, pages 3-55a and 3-55b, dated June 17, 2010, for related information. For service information related to this AD, contact Diamond Aircraft Industries GmbH, N.A. Otto-Straße 5, A-2700 Wiener Neustadt, Austria, telephone: +43 2622 26700; fax: +43 2622 26780; e-mail: [office@diamond-air.at](mailto:office@diamond-air.at); Internet: <http://www.diamond-air.at>. You may review copies of the referenced service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call 816-329-4148.

Issued in Kansas City, Missouri, on March 2, 2011.

**John R. Colomy,**

*Acting Manager, Small Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-5176 Filed 3-7-11; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. FAA-2011-0151; Directorate Identifier 2009-NM-205-AD]

RIN 2120-AA64

**Airworthiness Directives; Bombardier, Inc. Model DHC-8-400 Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for the products listed above that would supersede an existing AD. This proposed AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

Two cases of main landing gear collapse had been reported. Main landing gear collapse may result in unsafe landing of the aircraft.

\* \* \* \* \*

The proposed AD would require actions that are intended to address the unsafe condition described in the MCAI.

**DATES:** We must receive comments on this proposed AD by April 22, 2011.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* (202) 493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations,

M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

• **Hand Delivery:** U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-40, 1200 New Jersey Avenue, SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Bombardier, Inc., Q-Series Technical Help Desk, 123 Garratt Boulevard, Toronto, Ontario M3K 1Y5, Canada; telephone 416-375-4000; fax 416-375-4539; e-mail [thd.qseries@aero.bombardier.com](mailto:thd.qseries@aero.bombardier.com); Internet <http://www.bombardier.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

#### FOR FURTHER INFORMATION CONTACT:

Craig Yates, Aerospace Engineer, Airframe and Mechanical Systems Branch, ANE-171, FAA, New York Aircraft Certification Office (ACO), 1600 Stewart Avenue, Suite 410, Westbury, New York 11590; telephone (516) 228-7355; fax (516) 794-5531.

#### SUPPLEMENTARY INFORMATION:

##### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2011-0151; Directorate Identifier 2009-NM-205-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

[www.regulations.gov](http://www.regulations.gov), including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

#### Discussion

On October 19, 2007, we issued AD 2007-22-09, Amendment 39-15245 (72 FR 61288, October 30, 2007). That AD required actions intended to address an unsafe condition on the products listed above.

Since we issued AD 2007-22-09, Transport Canada Civil Aviation (TCCA), which is the aviation authority for Canada, has issued Canadian Airworthiness Directive CF-2007-20R2, dated February 6, 2009 (referred to after this as "the MCAI"), to correct an unsafe condition for the specified products. The MCAI states:

Two cases of main landing gear collapse had been reported. Main landing gear collapse may result in unsafe landing of the aircraft.

Revision 1 of this directive amended the time compliance in paragraph C.2 (3 months in addition to 500 hours air time), to add new paragraph C.3 to cater for retract actuator which has accumulated less than 4,000 landings or 2 years since new and to add new paragraphs B.2 and C.4 to require that the respective inspections be repetitively performed until terminating action becomes available.

Revision 2 of this directive amends the detailed visual inspection requirement in paragraph C.3 to include the main landing gear retract actuator, part number 46550-11, and to add new paragraph F to mandate the incorporation of main landing gear retract actuator part number, 46550-13 as the terminating action and to add new paragraph G for the maintenance requirement.

You may obtain further information by examining the MCAI in the AD docket.

#### Relevant Service Information

Bombardier has issued Service Bulletin 84-32-55, Revision A, dated March 10, 2008; and Repair Drawing 8/4-32-059, Issue 7, dated June 26, 2008. Bombardier has also issued Temporary Revision (TR) MRB-35, dated November 18, 2008, to Section 1-32 of Part 1 of the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7). The actions described in this service information are intended to correct the unsafe condition identified in the MCAI.

#### FAA's Determination and Requirements of This Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our

bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

#### Changes to AD 2007-22-09

This proposed AD would retain certain requirements of AD 2007-22-09. Since AD 2007-22-09 was issued, the AD format has been revised, and certain paragraphs have been rearranged. As a result, the corresponding paragraph identifiers have changed in this proposed AD, as listed in the following table:

#### REVISED PARAGRAPH IDENTIFIERS

Requirement in AD 2007-22-09	Corresponding requirement in this proposed AD
paragraph (f)	paragraph (g).
paragraph (g)	paragraph (h).
paragraph (h)	paragraph (i).
paragraph (i)	paragraph (j).
paragraph (j)	paragraph (k).
paragraph (l)	paragraph (v)(3).

We have revised paragraph (f) of AD 2007-22-09 to remove reference to Tasks Z700-03E and Z700-04E specified in Part 1 (Maintenance Review Board Report) of the Bombardier DHC-8 Series 400 Maintenance Requirements Manual (PSM 1-84-7). Instead, we have added Note 3 to this AD to specify that guidance on doing a general visual inspection to detect discrepancies of the left- and right-hand main landing gear system can be found in Tasks Z700-03E and Z700-04E of Part 1 (Maintenance Review Board Report) of the Bombardier DHC-8 Series 400 Maintenance Requirements Manual (PSM 1-84-7).

#### Change to Applicability of AD 2007-22-09

AD 2007-22-09 applies to airplanes having serial numbers (S/Ns) 003 and subsequent, which now corresponds to S/Ns 4003 and subsequent. This proposed AD applies to S/Ns 4003, 4004, 4006, and 4008 through 4208 inclusive, and also adds S/N 4001. The airplanes having serial numbers other than those specified in the Applicability of this proposed AD are not affected by the identified unsafe condition.

#### Differences Between This AD and the MCAI or Service Information

We have reviewed the MCAI and related service information and, in

general, agree with their substance. But we might have found it necessary to use different words from those in the MCAI to ensure the AD is clear for U.S. operators and is enforceable. In making these changes, we do not intend to differ substantively from the information provided in the MCAI and related service information.

We might also have proposed different actions in this AD from those in the MCAI in order to follow FAA policies. Any such differences are highlighted in a Note within the proposed AD.

#### Costs of Compliance

Based on the service information, we estimate that this proposed AD would affect about 55 products of U.S. registry.

The actions that are required by AD 2007–22–09 and retained in this proposed AD take about 5 work-hours per product, at an average labor rate of \$85 per work hour. Based on these figures, the estimated cost of the currently required actions is \$425 per product.

We estimate that it would take about 8 work-hours per product to comply with the new basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$27,511 per product. Where the service information lists required parts costs that are covered under warranty, we have assumed that there will be no charge for these costs. As we do not control warranty coverage for affected parties, some parties may incur costs higher than estimated here. Based on these figures, we estimate the cost of the proposed AD on U.S. operators to be \$1,550,505, or \$28,191 per product.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on

products identified in this rulemaking action.

#### Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
3. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

##### § 39.13 [Amended]

2. The FAA amends § 39.13 by removing Amendment 39–15245 (72 FR 61288, October 30, 2007) and adding the following new AD:

**Bombardier, Inc.:** Docket No. FAA–2011–0151; Directorate Identifier 2009–NM–205–AD.

#### Comments Due Date

- (a) We must receive comments by April 22, 2011.

#### Affected ADs

- (b) The AD supersedes AD 2007–22–09, Amendment 39–15245.

#### Applicability

- (c) This AD applies to Bombardier, Inc. Model DHC–8–400, –401, and –402 airplanes, certificated in any category, having serial numbers (S/Ns) 4001, 4003, 4004, 4006, and 4008 through 4208 inclusive.

#### Subject

- (d) Air Transport Association (ATA) of America Code 32: Landing Gear.

#### Reason

- (e) The mandatory continuing airworthiness information (MCAI) states: Two cases of main landing gear collapse had been reported. Main landing gear collapse may result in unsafe landing of the aircraft.

\* \* \* \* \*

#### Compliance

- (f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

#### Restatement of AD 2007–22–09, With Updated Service Information, Limited Affected Airplanes, and Revised Compliance Times

##### *General Visual Inspection of Main Landing Gear (MLG) System, and Corrective Actions*

(g) For airplanes having S/Ns 003, 004, 006, and 008 through 182 inclusive (now referred to as S/Ns 4003, 4004, 4006, and 4008 through 4182 inclusive), before further flight, do a general visual inspection to detect discrepancies of the left- and right-hand MLG system and do all applicable corrective actions, in accordance with a method approved by the Manager, New York Aircraft Certification Office (ACO), FAA; or Transport Canada Civil Aviation (TCCA) (or its delegated agent).

**Note 1:** For the purposes of this AD, a general visual inspection is: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to ensure visual access to all surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

**Note 2:** Guidance on doing a general visual inspection to detect discrepancies of the left- and right-hand MLG system can be found in Tasks Z700–03E and Z700–04E of Part 1 (Maintenance Review Board Report) of the Bombardier DHC–8 Series 400 Maintenance Requirements Manual (PSM 1–84–7).

#### General Visual Inspection of the Jam Nut of the Retract Actuator of the MLG and Corrective Actions

(h) For all airplanes except for the airplane having serial number 4001: Before further flight, do a general visual inspection of the jam nut of the retract actuator of the left- and right-hand MLG to ensure the wire lock is in place and the nut is secured. If the wire lock is not in place or if the jam nut is not secured, before further flight, adjust the retracted length of the rod end, torque the jam nut, install a wire lock, and lubricate the piston, as applicable, in accordance with

Bombardier Repair Drawing (RD) 8/4-32-059, Issue 4, dated September 14, 2007; or Issue 7, dated June 26, 2008. As of the effective date of this AD, use only Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008. Doing the revision required by paragraph (r) of this AD terminates the inspection required by this paragraph.

**Note 3:** Bombardier RD 8/4-32-059, Issue 4, dated September 14, 2007, refers to Goodrich Service Concession Request SCR 086-07, Revision C, dated September 14, 2007 (specifically item 14); and Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008, refers to Goodrich Service Concession Request SCR 086-07, Revision F, dated June 13, 2008 (specifically item 14); as an additional source of service information for adjusting the retracted length of the rod end, torquing the jam nut, installing a wire lock, and lubricating the piston if necessary, as required by paragraph (h) of this AD.

**Detailed Inspection of the Retract Actuator of the MLG, With Extended Compliance Time for Paragraph (j) of This AD**

(i) For airplanes having S/Ns 003, 004, 006, and 008 through 182 inclusive (now referred to as S/Ns 4003, 4004, 4006, and 4008 through 4182 inclusive) on which the retract actuator of the MLG, P/N 46550-7 or 46550-9, has accumulated 8,000 or more total

landings or has been in-service 4 or more years since new, as of November 14, 2007 (the effective date of 2007-22-09): Before further flight, do a detailed inspection of affected parts for any signs of corrosion or wear, and applicable related investigative and corrective actions, in accordance with Bombardier RD 8/4-32-059, Issue 4, dated September 14, 2007; or Issue 7, dated June 26, 2008. As of the effective date of this AD, use only Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008.

**Note 4:** For the purposes of this AD, a detailed inspection is: "An intensive examination of a specific item, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at an intensity deemed appropriate. Inspection aids such as mirror, magnifying lenses, etc., may be necessary. Surface cleaning and elaborate procedures may be required."

(j) For airplanes having S/Ns 003, 004, 006, and 008 through 182 inclusive (now referred to as S/Ns 4003, 4004, 4006, and 4008 through 4182 inclusive) with a retract actuator of the MLG, P/N 46550-7 or 46550-9, other than those identified in paragraph (i) of this AD: Do a detailed inspection of affected parts for any signs of corrosion or wear, and applicable related investigative

and corrective actions, in accordance with Bombardier RD 8/4-32-059, Issue 4, dated September 14, 2007; or Issue 7, dated June 26, 2008; at the later of the times specified in paragraphs (j)(1) and (j)(2) of this AD. As of the effective date of this AD, use only Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008.

(1) Before the accumulation of 4,500 total landings or 27 months since new, whichever occurs first.

(2) Within 500 flight hours after November 14, 2007, or within 3 months after the effective date of this AD, whichever occurs first.

**Note 5:** Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008, refers to Goodrich Service Concession Request SCR 086-07, Revision F, dated June 13, 2008, as an additional source of service information for accomplishing the applicable related investigative and corrective actions required by paragraphs (i) and (j) of this AD.

**Actions Done in Accordance With Previous Issues of Service Information**

(k) Actions done before November 14, 2007, in accordance with repair drawings specified in Table 1 of this AD, are acceptable for compliance with the corresponding actions specified in paragraphs (h) through (j) of this AD.

TABLE 1—PREVIOUS REPAIR DRAWINGS

Document	Issue	Date
Bombardier Repair Drawing 8/4-32-059 .....	1	September 12, 2007.
Bombardier Repair Drawing 8/4-32-059 .....	2	September 13, 2007.
Bombardier Repair Drawing 8/4-32-059 .....	3	September 13, 2007.

**New Requirements of this AD**

*General Visual Inspection of the Jam Nut of the Retract Actuator of the MLG, and Corrective Actions*

(l) For all airplanes: At the later of the times specified in paragraphs (l)(1) and (l)(2) of this AD, do a general visual inspection of the left- and right-hand MLG retract actuator jam nut to ensure that the wire lock is in place and that the nut is secure, in accordance with a method approved by the Manager, New York Aircraft Certification Office (ACO), FAA; or Transport Canada Civil Aviation (TCCA) (or its delegated agent). If the wire lock is not in place or the jam nut is not secured, before further flight, re-torque the jam nut and safety lockwire, in accordance with Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008. Repeat the inspection thereafter at intervals not to exceed 250 flight cycles or 30 days, whichever occurs first. Doing the revision required by paragraph (r) of this AD terminates the inspections required by this paragraph.

(1) Within 250 flight cycles or 30 days after accomplishing the inspection required by paragraph (h) of this AD, whichever occurs first.

(2) Within 7 days after the effective date of this AD.

**Note 6:** Guidance for doing a general visual inspection to detect discrepancies of the left- and right-hand MLG system can be found in Tasks Z700-03E and Z700-04E of Part 1 (Maintenance Review Board Report) of the Bombardier DHC-8 Series 400 Maintenance Requirements Manual (PSM 1-84-7).

**Detailed Inspection of the Retract Actuator of the MLG, and Related Investigative and Corrective Actions**

(m) For airplanes equipped with a MLG retract actuator having P/N 46550-7 or 46550-9: At the later of the times specified in paragraphs (m)(1) and (m)(2) of this AD, do a detailed inspection of affected parts for any signs of corrosion or wear, and do applicable related investigative and corrective actions, in accordance with Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008. Do all applicable related investigative and corrective actions before further flight. Repeat the inspection thereafter at intervals not to exceed 2,000 flight cycles or 12 months, whichever occurs first.

(1) Within 2,000 flight cycles or within 12 months after accomplishing the inspection required by paragraph (i) or (j) of this AD, whichever occurs first.

(2) Within 30 days after the effective date of this AD.

(n) For airplanes having serial numbers 4001, 4003, 4004, 4006, and 4008 through 4182 inclusive equipped with a MLG retract actuator having P/N 46550-11: At the later of the times specified in paragraphs (n)(1) and (n)(2) of this AD, do a detailed inspection of affected parts for any signs of corrosion or wear, and applicable related investigative and corrective actions, in accordance with Bombardier RD 8/4-32-059, Issue 7, dated June 26, 2008. Do all applicable related investigative and corrective actions before further flight. Repeat the inspection thereafter at intervals not to exceed 2,000 flight cycles or 12 months, whichever occurs first.

(1) Before the accumulation of 4,500 total landings or 27 months since new, whichever occurs first.

(2) Within 500 flight hours or 3 months after the effective date of this AD, whichever occurs first.

(o) For airplanes having serial numbers 4001, 4003, 4004, 4006, and 4008 through 4182 inclusive equipped with a MLG retract actuator having P/N 46550-7, P/N 46550-9, or P/N 46550-11, and that have accumulated 7,500 total flight cycles or more as of the effective date of this AD, or that have more than 48 months since new: Within 500 flight cycles or 3 months after the effective date of this AD, whichever occurs first, replace the affected retract actuator with a new design

retract actuator having P/N 46550-13, in accordance with Bombardier Service Bulletin 84-32-55, Revision A, dated March 10, 2008 (Bombardier Modsum 4-901603). Doing the replacement specified in this paragraph terminates the requirements of paragraphs (i), (j), (m), and (n) of this AD.

(p) For airplanes having serial numbers 4001, 4003, 4004, 4006, and 4008 through 4182 inclusive equipped with MLG retract actuators having P/N 46550-7, P/N 46550-9, or P/N 46550-11, that have accumulated less than 7,500 total flight cycles as of the effective date of this AD and that have 48 months or less since new: Prior to the accumulation of 8,000 total flight cycles, or within 51 months since new, whichever occurs first, replace the affected retract actuator with a new design retract actuator having P/N 46550-13, in accordance with Bombardier Service Bulletin 84-32-55, Revision A, dated March 10, 2008 (Bombardier Modsum 4-901603). Doing the replacement specified in this paragraph terminates the requirements of paragraphs (i), (j), (m), and (n) of this AD.

(q) Replacing the affected retract actuator with a new design retract actuator having P/N 46550-15, in accordance with the Accomplishment Instructions of Bombardier Service Bulletin 84-32-60, Revision A, dated September 29, 2008 (Bombardier Modsum 4-901610), is also acceptable for compliance with the requirements of paragraphs (o) and (p) of this AD.

#### Revision of the Maintenance Program

(r) For all airplanes: Within 30 days after the effective date of this AD, revise the maintenance program by incorporating Task 320100-211 (repetitive detailed inspections of the retraction actuator rod end jam nut, gland nut, and actuator attachment pins for condition, the security of installation, and corrosion) and Task 320100-212 (repetitive restorations of the retraction actuator for complete overhaul), as specified in Bombardier Temporary Revision (TR) MRB-35, dated November 18, 2008, to the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7). Doing this revision terminates the requirements of paragraphs (h) and (l) of this AD. The initial compliance times for doing Task 320100-211 and Task 320100-212 are specified in paragraphs (r)(1) and (r)(2) of this AD. After doing this revision, no alternative inspections, restorations, or intervals may be used, unless the inspections, restorations, or intervals are approved as an alternative method of compliance (AMOC) in accordance with the procedures specified in paragraph (v)(1) of this AD.

(1) For Task 320100-211 in Bombardier TR MRB-35, dated November 18, 2008, to the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7): The compliance time for the initial inspection is within 600 flight hours after the effective date of this AD.

(2) For Task 320100-212 in Bombardier TR MRB-35, dated November 18, 2008, to the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7): The compliance time for the initial restoration is the later of the times of paragraphs (r)(2)(i) and (r)(2)(ii) of this AD.

(i) Prior to the accumulation of 25,000 total flight cycles, or within 12 years since the date of issuance of the original airworthiness certificate or the date of issuance of the original export certificate of airworthiness, whichever occurs first.

(ii) Within 500 flight cycles after the effective date of this AD.

**Note 7:** The actions required by paragraph (r) of this AD may be done by inserting copies of Bombardier TR MRB-35, dated November 18, 2008, into the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7). When this TR has been included in general revisions of the PSM, the general revisions may be inserted in the PSM, provided the relevant information in the general revision is identical to that in Bombardier TR MRB-35, dated November 18, 2008.

#### Credit for Actions Accomplished in Accordance With Previous Service Information

(s) Doing a general visual inspection of the jam nut of the retract actuator of the left- and right-hand MLG; and doing a detailed inspection of affected parts for any signs of corrosion or wear, and applicable related investigative and corrective actions; is also acceptable for compliance with the corresponding requirements of paragraphs (h), (i), (j), (l), (m), and (n) of this AD, if done before the effective date of this AD in accordance with Bombardier Repair Drawing 8/4-32-059, Issue 5, dated September 20, 2007; or Bombardier Repair Drawing 8/4-32-059, Issue 6, dated January 31, 2008.

(t) Replacing the affected retract actuator with a new design retract actuator having P/N 46550-13 is also acceptable for compliance with the requirements of paragraphs (o) and (p) of this AD, if done before the effective date of this AD in accordance with Bombardier Service Bulletin 84-32-55, dated January 14, 2008 (Modsum 4-901603).

#### No Reporting

(u) While Canadian Airworthiness Directive CF-2007-20R2, dated February 6, 2009, has a reporting action, this AD does not require reporting.

#### FAA AD Differences

**Note 8:** This AD differs from the MCAI and/or service information as follows: Although the MCAI or service information tells you to submit information to the manufacturer, paragraph (u) of this AD specifies that such submittal is not required.

#### Other FAA AD Provisions

(v) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, New York Aircraft Certification Office (ACO), ANE-170, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Program Manager, Continuing Operational Safety, FAA, New York ACO, 1600 Stewart Avenue, Suite 410, Westbury, New York, 11590; telephone 516-228-7300; fax 516-794-5531. Before using any approved AMOC

on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically reference this AD. AMOCs approved previously in accordance with AD 2007-22-09, Amendment 39-15245, are approved as AMOCs for the corresponding provisions of paragraph (i) and (j) of this AD.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Special Flight Permits:* Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the airplane can be inspected (if the operator elects to do so), provided that the procedures and limitations in paragraphs (v)(3)(i) and (v)(3)(ii) of this AD are adhered to.

(i) Flight Crew Limitations and Procedures:  
(A) Ferry flight with gear extended and pinned;

(B) Landing to be conducted at a minimum descent rate;

(C) Minimize braking on landing;

(D) Flight to be conducted in accordance with Section 4.8 of the Aircraft Operating Manual (AOM);

(E) Only essential crew on board; and

(F) Flight in known or forecast icing condition is prohibited.

(ii) Maintenance Procedures:

(A) Do the general visual inspection required by paragraph (h) of this AD;

(B) Do the general visual inspections of the stabilizer stay and the hinge points of the MLG for general condition and security, in accordance with Bombardier Q400 All Operator Message 236A, dated September 11, 2007;

(C) If no discrepancy is detected during the inspections required by paragraph (v)(3)(ii)(A) and (v)(3)(ii)(B) of this AD, before further flight, insert the ground lock pins and a wire lock of the MLG in place.

(D) Ensure the nose landing gear ground lock is engaged.

#### Related Information

(w) Refer to MCAI Canadian Airworthiness Directive CF-2007-20R2, dated February 6, 2009; Bombardier Service Bulletin 84-32-55, Revision A, dated March 10, 2008; Bombardier Service Bulletin 84-32-60, Revision A, dated September 29, 2008; Bombardier Repair Drawing 8/4-32-059, Issue 7, dated June 26, 2008; and Bombardier Temporary Revision MRB-35, dated November 18, 2008, to the Bombardier Q400 Dash 8 Maintenance Requirements Manual (PSM 1-84-7); for related information.

Issued in Renton, Washington, on February 22, 2011.

**Kalene C. Yanamura,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2011-5161 Filed 3-7-11; 8:45 am]

BILLING CODE 4910-13-P

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2007-28661; Directorate Identifier 2007-NM-013-AD]

RIN 2120-AA64

#### **Airworthiness Directives; The Boeing Company Model 737-600, -700, -700C, -800, and -900 Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Supplemental notice of proposed rulemaking (NPRM); reopening of comment period.

**SUMMARY:** We are revising an earlier proposed airworthiness directive (AD) for the products listed above. That NPRM proposed to require installation of an automatic shutoff system for the center tank fuel boost pumps, installation of a placard in the airplane flight deck if necessary, and concurrent modification of the P5-2 fuel control module assembly. That NPRM also proposed to require revisions to the Limitations and Normal Procedures sections of the airplane flight manual to advise the flightcrew of certain operating restrictions for airplanes equipped with an automated center tank fuel pump shutoff control. Additionally, that NPRM proposed to require a revision to the Airworthiness Limitations (AWL) section of the Instructions for Continued Airworthiness (ICA) to incorporate AWL No. 28-AWL-19 and No. 28-AWL-23. That NPRM further proposed to require installation of a secondary control relay for the electrical control circuit of each of the two center tank fuel boost pumps. That NPRM was prompted by fuel system reviews conducted by the manufacturer. This action revises that NPRM by adding airplanes, adding additional operational testing of the automatic shutoff system for certain airplanes, removing the requirement for incorporating AWL No. 28-AWL-19 into the AWL section of the ICA, and adding an option of installation and maintenance of universal fault interrupters using a certain supplemental type certificate. We are proposing this supplemental NPRM to

prevent center tank fuel pump operation with continuous low pressure, which could lead to friction sparks or overheating in the fuel pump inlet that could create a potential ignition source inside the center fuel tank. These conditions, in combination with flammable fuel vapors, could result in a center fuel tank explosion and consequent loss of the airplane. Since these actions impose an additional burden over those proposed in the NPRM, we are reopening the comment period to allow the public the chance to comment on these proposed changes.

**DATES:** We must receive comments on this supplemental NPRM by April 4, 2011.

**ADDRESSES:** You may send comments by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.
- *Hand Delivery:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H-65, Seattle, Washington 98124-2207; telephone 206-544-5000, extension 1; fax 206-766-5680; e-mail [me.boecom@boeing.com](mailto:me.boecom@boeing.com); Internet <https://www.myboeingfleet.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221.

#### **Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Tak Kobayashi, Aerospace Engineer, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057-3356; *phone:* (425) 917-6499; *fax:* (425) 917-6590; *e-mail:* [Takahisa.Kobayashi@faa.gov](mailto:Takahisa.Kobayashi@faa.gov).

#### **SUPPLEMENTARY INFORMATION:**

##### **Comments Invited**

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2007-28661; Directorate Identifier 2007-NM-013-AD;" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

##### **Discussion**

We issued an NPRM to amend 14 CFR part 39 to include an airworthiness directive (AD) that would apply to certain Model 737-600, -700, -700C, -800, and -900 series airplanes. That NPRM published in the **Federal Register** on July 10, 2007 (72 FR 37479). That NPRM proposed to require installation of an automatic shutoff system for the center tank fuel boost pumps, installation of a placard in the airplane flight deck if necessary, and concurrent modification of the P5-2 fuel control module assembly. That NPRM proposed to require revisions to the Limitations and Normal Procedures sections of the airplane flight manual to advise the flightcrew of certain operating restrictions for airplanes equipped with an automated center tank fuel pump shutoff control. Additionally, that NPRM proposed to require a revision to the Airworthiness Limitations (AWL) section of the Instructions for Continued Airworthiness (ICA) to incorporate AWL No. 28-AWL-19 and No. 28-AWL-23. That NPRM also proposed to require installation of a secondary control relay for the electrical control circuit of each of the two center tank fuel boost pumps.